

Fig. 3 Incipient separation pressure rise.

Conclusions

Although two-dimensional data cover a wide band of incipient separation angles θ_i , it is clear that the values are considerably higher than for skewed shock wave interactions, and the gap widens with increasing Mach number. Thus, it is the skewed shock wave interaction with the sidewall turbulent boundary layer in rectangular diffusers or inlets (see Fig. 1) that first leads to separation and possible flow breakdown for compression angles (or pressure rises) which may be well below the incipient values for the two-dimensional case.

References

¹ Roshko, A. and Thomke, G. J., "Supersonic, Turbulent Boundary Layer Interaction with a Compression Corner at Very High Reynolds Numbers," Proceedings of the Symposium on Viscous Interaction Phenomena in Supersonic and Hypersonic Flow, Aerospace Research Labs., May 1969, pp. 109–138.

² Law, C. H., "Supersonic, Turbulent Boundary Layer Separation,"

AIAA Journal, Vol. 12, No. 6, June 1974, pp. 794-797.

³ Coleman, G. T., Elfstrom, G. M., and Stollery, J. L., "Turbulent Boundary Layers at Supersonic and Hypersonic Speeds," AGARD Conference Proceedings, No. 93 on "Turbulent Shear Flows," Paper

31, Sept. 1971.

⁴ Korkegi, R. H., "Survey of Viscous Interactions Associated with High Mach Number Flight," *AIAA Journal*, Vol. 9, No. 5, May 1971,

pp. 771-784.

⁵ Korkegi, R. H., "A Simple Correlation for Incipient Turbulent Boundary Layer Separation Due to a Skewed Shock Wave," AIAA Journal, Vol. 11, No. 11, Nov. 1973, pp. 1578-1579.

⁶ Drougge, G., "An Experimental Investigation of the Influence of Strong Adverse Pressure Gradients on Turbulent Boundary Layers at

Supersonic Speeds," FFA Rept. 47, The Aeronautical Research Institute of Sweden, Stockholm, Sweden, 1953.

Kuehn, D. M., "Experimental Investigation of the Pressure Rise Required for the Incipient Separation of Turbulent Boundary Layers in Two-Dimensional Supersonic Flow," Memo 1-21-59A, 1959, NASA.

⁸ Settles, G. S. and Bogdonoff, S. M., "Separation of a Supersonic

Turbulent Boundary Layer at Moderate to High Reynolds Numbers,"

AIAA Paper 73-666, Palm Springs, Calif., 1973.

Sterret, J. R. and Emery, J. C., "Experimental Separation Studies for Two-Dimensional Wedges and Curved Surfaces at M = 4.8 to 6.2, TN D-1014, 1962, NASA.

¹⁰ Gary, J. D. and Rhudy, R. W., "Investigation of Flat-Plate Aspect Ratio Effects on Ramp-Induced, Adiabatic, Boundary Layer Separation at Supersonic and Hypersonic Speeds," AEDC-TR-70(235), March 1971, Arnold Engineering Development Center, Tullahoma, Tenn.

Batham, J. P., "An Experimental Study of Turbulent Separating and Reattaching Flows at a High Mach Number," Journal of Fluid

Mechanics, Vol. 52, Pt. 3, April 1972, pp. 425-435.

¹² Elfstrom, G. M., "Turbulent Hypersonic Flow at a Wedge-Compression Corner," *Journal of Fluid Mechanics*, Vol. 53, Pt. 1, May

¹³ Holden, M. S., "Shock Wave-Turbulent Boundary Layer Interaction in Hypersonic Flow," AIAA Paper 72-74, San Diego, Calif.,

1972.

14 Neumann, R. D. and Token, K. H., Prediction of Surface Phenomena Induced by Three-Dimensional Interactions on Planar Turbulent Boundary Layers," Paper 74-058, International Astronautical Federation XXV Congress, Amsterdam, The Netherlands,

Oct. 1974.

15 Law, C. H., unpublished data, Aug. 1974, Aerospace Research Labs., Wright-Patterson Air Force Base, Ohio.

¹⁶ Goldberg, J. J., "Three-Dimensional Separation for Interaction of Shock Waves with Turbulent Boundary Layers," AIAA Journal, Vol. 11, No. 11, Nov. 1973, pp. 1573-1575.

Interferometric Technique for Measuring Mixing of a **Buoyant Plume**

ARVEL B. WITTE* AND DAVID D. MANTROM† TRW Systems Group, Redondo Beach, Calif.

WHEN investigating the three-dimensional and unstable character of buoyant plumes, it is important to use a measurement technique which does not disturb the flowfield. Interferometry provides a simple means of making detailed concentration measurements under conditions which have a range of applicability. This Note concerns itself with a special case, namely, the isothermal, isobaric plume. A simple development follows which shows how interferometry can be used to measure buoyant plume mixing of a light gas into a heavy background gas, all at constant temperature and pressure. Finally, an example of the results is given in the form of reduced data of a vortex-like plume.

The standard equation for fringe shift is:

$$S = 1/\lambda \int_0^L (n - n_\infty) ds \tag{1}$$

where S = fringe number, $\lambda =$ wave length of light in vacuum, L = integration path length, n = index of refraction, s = pathlength along the light ray, and subscript ∞ refers to reference conditions.

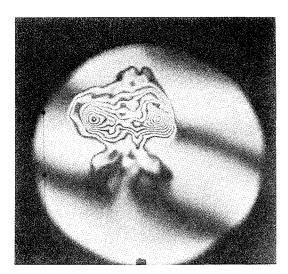
An interferogram yields the fringe number S, and the solution of the integral equation provides the index field, n. An equation for n can be developed in terms of the species mass fraction C_i and Gladstone-Dale constants K_i as

Received August 8, 1974; revision received September 23, 1974. Index categories: Jets, Wakes and Viscid-Inviscid Flow Interactions;

Shock Waves and Detonations. * Manager, Fluid Physics Department, TRW Systems Group,

Redondo Beach, Calif.

† Presently a Member of the Technical Staff, Techniate, Inc., Torrance, Calif.



Holographic interferogram of vortex formed buoyant rise of bursted helium-filled soap bubble in air (4 atm pressure).

 $\frac{n - n_H}{\rho_H K_H} = \frac{\rho}{\rho_H} \left[C_L \left(\frac{K_L}{K_H} - 1 \right) + 1 \right] - 1$ (2)

where

$$n = \sum K_i \rho_i + 1 = \rho \sum K_i C_i + 1$$
$$C_L + C_H = 1$$
$$n_{\infty} = n_H$$

were used. The subscripts H and L refer to heavy and light molecular weight gas conditions, respectively.

The density ratio:

$$\frac{\rho}{\rho_H} = \left[C_L \left(\frac{M_H}{M_L} - 1 \right) + 1 \right]^{-1} \tag{3}$$

follows from the perfect gas la

$$\rho/\rho_H = (p/p_H)(T_H/T)(M/M_H)$$

the assumptions of constant pressure and temperature, $p = p_H$ and $T = T_H$ and the definition of average molecular weight M,

$$1/M = C_L(1/M_L - 1/M_H) + 1/M_H$$

developed from $1/M = \sum C_i/M_i$. By eliminating ρ/ρ_H between Eqs. (2) and (3), one obtains the following expression for the light species mass fraction,

$$C_{L} = \frac{(n_{H} - n)/\rho_{H} K_{H}}{\{[(n - n_{H})/\rho_{H} K_{H}] + 1\}[(M_{H}/M_{L}) - 1] - [(K_{L}/K_{H}) - 1]}$$
(4)

The solution of any problem commences with the solution of an integral equation, Eq. (1), for the quantity $n-n_H$. For the general three-dimensional index field, interferometric data must be available for various angular views of the phenomenon as formulated by Witte¹ and implemented in detail by Matulka.² This can be accomplished in principle by recording a holographic interferogram having 180° of angular viewing or by recording a series of interferograms by rotating the phenomenon about the test section of the interferometer. When the phenomenon is axisymmetric, the Abel inversion integral is used to solve for nas a function of S.³

An example of the technique is provided by solving for the species mass fraction of a helium (or nitrogen) plume rising in air (or sulfur-hexafluoride). A typical interferogram is shown in Fig. 1. Initially, a spherical helium filled soap bubble is burst in air. A vortex-like plume forms after a rise of about 4 initial bubble diameters. The vortical motion quickly entrains air, especially into the region near the vertical axis of rise. Fringe data are read from the interferogram by counting fringes along slices of the plume taken perpendicular to the axis. The first dark fringe has an absolute value of $\frac{1}{2}$ and is negative relative to the air background because the optical path decreases as one proceeds into the helium plume. Because of statistical variations in these flow phenomena, a data-averaging scheme needs to be developed to

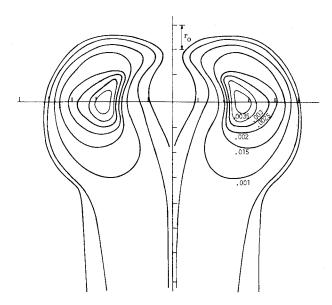


Fig. 2 ISO-concentration (N_2 mass fraction) profiles for a turbulent vortex rising in SF_6 . Initial conditions: N_2 filled soap bubble, SF_6 background, $r_0 = \frac{1}{4}$ in. (initial bubble radius). Profile data = $h/2r_0 = 10.4$ (height of phenomena), 150 profiles denote N_2 mass fraction.

arrive at the correct mean flowfield. Such an averaging scheme was developed and applied to these data by Mantrom and Haigh.4 Data reduction provides the results in Fig. 2 for the isoconcentration profiles (N_2 mass fraction) for a turbulent vortex formed by bursting an N2-filled bubble into a 10-atm SF6 background. The initial bubble diameter was $\frac{1}{2}$ in. and the data were recorded at a height of ~ 10.4 initial bubble diameters. Mixing with the background gas is observed to be all but complete; however, the contours still show the vortex shape. Further details of these data as well as those for the helium vortex phenomena are given in Ref. 4.

References

¹ Witte, A. B., "Three-Dimensional Flow Field Analysis by Holography and Interferometry," TRW Rept. 12414-6003-R0-00, Nov.

² Matulka, R. D. and Collins, D. J., "Determination of Three-Dimensional Density Fields from Holographic Interferograms," Journal of Applied Physics, Vol. 42, No. 3, March 1971.

³ Ladenburg, R. W., "Physical Measurements in Gas Dynamics and Combustion," High Speed Aerodynamics and Jet Propulsion, Pt. 1, Vol. IX, Princeton University Press, Princeton, N.J.

⁴ Mantrom, D. D. and Haigh, W. W., "Fireball Entrainment Study," TRW Rept. 18895-6004-RV-00, DNA 2981Z, Sept. 1973.

Two-Dimensional Supersonic Diffuser Experiments

SIEGFRIED H. HASINGER* AND DAVID K. MILLER† Aerospace Research Laboratories, Wright-Patterson Air Force Base, Ohio

THE design of supersonic diffusers, as used in supersonic wind tunnels is well understood. The flow mechanism primarily relied upon is that of the so-called "pseudo shock,"

Received August 23, 1974; revision received September 26, 1974. Index categories: Nozzle and Channel Flow; Supersonic and Hypersonic Flow

* Research Scientist, Energy Conversion Research Laboratory. Member AIAA.

† Group Leader, Energy Conversion Research Laboratory, Capt., U.S. Air Force.